

IN THE CLAIMS:

1-6. (canceled)

7. (currently amended) A primer nozzle for a gas turbine engine combustor including a centerline axis, said primer nozzle comprising:

an inlet coupled to a source of pressurized air;

an injection tip for discharging fuel into said combustor in a direction that is substantially parallel to the gas turbine engine centerline axis;

a body extending between said inlet and said injection tip, said body comprising at least one annular projection for coupling said nozzle to said body such that said primer nozzle is positioned relative to the combustor; and

a shroud extending around said injection tip and around at least a portion of said body such that a gap is defined between said shroud and at least one of said body and said injection tip, said shroud comprising a shroud tip extending around said injection tip, said shroud tip comprising a plurality of cooling openings extending therethrough to facilitate film cooling said injection tip, said shroud comprising a plurality of circumferentially-spaced openings for metering cooling air supplied to said injection tip, wherein said plurality of circumferentially-spaced openings are coupled in flow communication to a recuperator for receiving cooling air therefrom.

8. (original) A primer nozzle in accordance with Claim 7 wherein said primer nozzle configured to supply fuel to the gas turbine engine combustor only during engine start-up operating conditions.

9. (canceled)

10. (original) A primer nozzle in accordance with Claim 7 wherein said shroud further comprises a shroud tip extending around said injection tip, said shroud tip is frusto-conical.

11. (original) A primer nozzle in accordance with Claim 10 wherein said shroud plurality of circumferentially-spaced openings facilitate limiting an airflow therethrough if said shroud tip deteriorates.

12. (previously presented) A primer nozzle in accordance with Claim 7 wherein said source of pressurized air comprises an accumulator, said inlet for channeling air from said air source for purging residual fuel into the combustor from said nozzle during pre-determined combustor operating conditions.

13. (previously presented) A combustion system for a gas turbine engine, said combustion system comprising:

a combustor comprising a dome assembly and a combustor liner extending downstream from said dome assembly, said combustor liner defining a combustion chamber therein, said combustor further comprising a centerline axis;

a combustor casing extending around said combustor;

an annular support ring comprising a first radial flange, a second radial flange axially spaced from said first radial flange, and a plurality of circumferentially-spaced beams extending between said first radial flange and said second radial flange, said combustor casing coupled to said annular support ring; and

a primer nozzle extending axially through said annular support ring, said combustor casing, and said dome assembly for supplying fuel into said combustor along said combustor centerline axis during engine start-up operating conditions, said primer nozzle comprising an inlet coupled to a source of pressurized air.

14. (canceled)

15. (original) A combustion system in accordance with Claim 13 wherein said primer nozzle comprises an annular shoulder, said primer nozzle positioned relative to said combustor casing by said shoulder.

16. (previously presented) A combustion system in accordance with Claim 15 wherein said primer nozzle comprises an injection tip and a body extending therebetween,

said injection tip for discharging fuel into said combustor in a direction that is substantially parallel to said combustor centerline axis.

17. (previously presented) A combustion system in accordance with Claim 13 wherein said primer nozzle comprises an injection tip, a body extending between said tip and said inlet, and a shroud extending circumferentially around said injection tip and around at least a portion of said body such that a gap is defined between said shroud and at least one of said body and said injection tip.

18. (original) A combustion system in accordance with Claim 17 wherein said shroud comprises a plurality of circumferentially-spaced metering openings extending therethrough, said metering openings for metering a flow of cooling air to said injection tip.

19. (original) A combustion system in accordance with Claim 17 wherein said shroud comprises a frusto-conical tip.

20. (previously presented) A combustion system in accordance with Claim 13 wherein said source of pressurized air comprises an accumulator, said inlet for channeling air from said air source is used for purging residual fuel into the combustor from said primer nozzle during pre-determined combustor operating conditions.